A Novel Wireless Enabled SHM-RFID-IoT Smart Fatigue Damage Sensor and Wireless SHM-RFID-IoT Smart Fatigue Damage SENSOR NETWORK and **An Internet of Things-IoT Based Intelligent Predictive Maintenance Management System**

(FOR AIRCRAFT APPLICATIONS)



STRUCTURAL FATIGUE DAMAGE **MONITORING** of different components of AIRCRAFT **SYSTEMS** (body frames or fatigue specific-sensitive elements) during normal service.

FATIGUE SENSOR APPLICATIONS



Identify all the Fatigue Critical HOT SPOTS in STRUCTURES for the applications of Smart fatigue damage sensors in order to monitor the RESIDUAL FATIGUE **STRENGTH or Fatigue Lifetime. Specific and Fatigue sensitive** regions, locations under high loads, predetermined and formerly known-experienced spots on the structures and mechanical components such as Riveted, **Bolted and Hole Type Connections** etc...

Wireless Enabled SHM-RFID-IoT Smart Fatigue Damage Sensor Network

(WIRELESS FATIGUE DAMAGE SENSOR NETWORK FOR INTELLIGENT STRUCTURAL HEALTH MONITORING, MAINTENANCE AND DESIGN)



Fatigue sensitive regions, locations under high loads, predetermined and formerly knownexperienced spots on the structures and mechanical components such as WELDED Connections etc..



A Novel Wireless Enabled SHM-RFID-IoT Smart Fatigue Damage Sensor and SHM STRUCTURAL HEALTH MONITORING OF AIRCRAFTS

Widespread Fatigue Damage Detection



STRUCTURAL FATIGUE DAMAGE MONITORING AND FATIGUE DAMAGE **DETECTION** of different components of AIRCRAFT SYSTEMS (body frames or fatigue specific-sensitive elements) during normal service.



WIRELESS STRUCTURAL FATIGUE DAMAGE MONITORING AND FATIGUE DAMAGE DETECTION of different components of **AIRCRAFT SYSTEMS** (body frames or fatigue specificsensitive elements) during normal service.



WIRELESS STRUCTURAL FATIGUE DAMAGE MONITORING AND FATIGUE DAMAGE DETECTION-SYSTEM MODEL (AIRCRAFT)

Fatigue sensitive regions, locations under high loads, predetermined and formerly knownexperienced spots on the structures and mechanical components such as Riveted, Bolted and Hole Type Connections etc..



CRACK







A schematic of the Hawk tailplane (upper left) and its detail (upper right) for VIT's fracture mechanics analyses: An element model of the detail (center left), estimated stress intensity factor (K) values as the function of the distance from the hole edge using various solutions, e.g. [Harter 2000; FRANC2D/L] (lower right). The simplified fatigue test specimen of the tailplane butt strap tested by HUT/LLS (bottom left), which was analyzed by PFA (bottom right).

Local Approaches base on Fracture Mechanics or based of S-N of ε-N curves of plain material may be used as a true alternative.

Local Approaches requires detailed stress analysis of riveted joints.



J. Correia & A. Jesus, 7ICSB - SteelBridges, 2008.







APPLICATIONS OF THE SENSOR TO AIRCRAFT WINGS:











FATIGUE FRACTURE OF A MAIN LANDING GEAR SWINGING LEVER IN A CIVIL AIRCRAFT



Submitted to-Prof. R.K.Pandey Submitted By Pramod Kawade 2012AMD122613

Introduction

 The left-hand main landing gear (LH-MLG) of a civil aircraft collapsed during the takeoff due to the fracture of its swinging lever.







